

CLAIMS

1. A disk of a disk rotor for a gas turbine comprising a central portion (22), an intermediate portion (24), an outer portion (28), a series of axial pass-through holes (27) for a series of tie rods and a series of slots (50) for housing a corresponding series of vanes, said central portion (22) comprises a central axial pass-through hole (23), a first collar (30) situated at a first end and a second collar (40) situated at a second end of the central portion (22), characterized in that the series of holes (27) is positioned in the outer portion (28) of the disk so as to obtain high dynamic characteristics of the rotor and at the same time a sufficient useful life thereof.
2. The disk of a disk rotor for a gas turbine according to claim 1, characterized in that said series of holes (27) is situated on a base surface (31) of the outer portion (28).
3. The disk of a disk rotor for a gas turbine according to claim 2, characterized in that the holes of said series of holes (27) are positioned at an equal distance from each other along a circumference (61) lying on the base surface (31), said circumference (61) is coaxial with the axis of the disk.
4. The disk of a disk rotor for a gas turbine according

to claim 1, characterized in that said first collar (30) comprises a bevel (38) and a relief (36) and in that said second collar (40) comprises a bevel (48) and a relief (46).

5 5. The disk of a disk rotor for a gas turbine according to claim 1, characterized in that the disk (20) has a total number of holes (27) of the series of holes which is equal to the total number of slots (50) of the series of slots (50) for the series of vanes.

10 6. The disk of a disk rotor for a gas turbine according to any of the previous claims, a point (80) having been defined for each slot, obtained by the intersection of an axis of the slot (82) in the middle side section of the disk (20) with the extension of the side surface (29),
15 and an angle (83) having been defined, which indicates the angular reference between the centre of a hole (27) and the position of the point (80) of an adjacent slot, said disk being characterized in that said angle (83) ranges from 2 to 10 sexagesimal degrees.

20 7. The disk of a disk rotor for a gas turbine according to claim 6, characterized in that said angle (83) ranges from 4 to 8 sexagesimal degrees.

8. The disk of a disk rotor for a gas turbine according to any of the previous claims, characterized in that the
25 diameter of the circumference (61) is close to the diame-

ter of the disk (20).

9. A disk rotor for a compressor comprising a series of disks (20) according to any of the previous claims and also comprising a series of tie rods, and a series of
5 vanes.

10. A disk of a disk rotor for a gas turbine as previously described and illustrated and for the purposes specified above.

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